



## Co-simulation modelling of a medium sized thermal vacuum facility for test feasibility studies



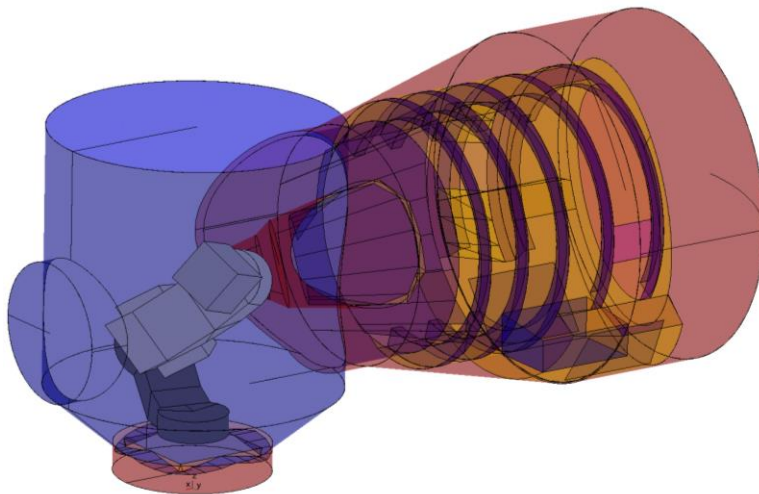
The European Space Agency

Presented By  
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**TFAWS**  
MSFC • 2017

Thermal & Fluids Analysis Workshop  
TFAWS 2017  
August 21-25, 2017  
NASA Marshall Space Flight Center  
Huntsville, AL

- Thermal Modelling of TVAC chambers (LSS, PHENIX)
- Spacecraft thermal design:
  - Proba 3 / SMILE / JUICE
- STEP-TAS model conversions
- Development of thermal tools – post processor

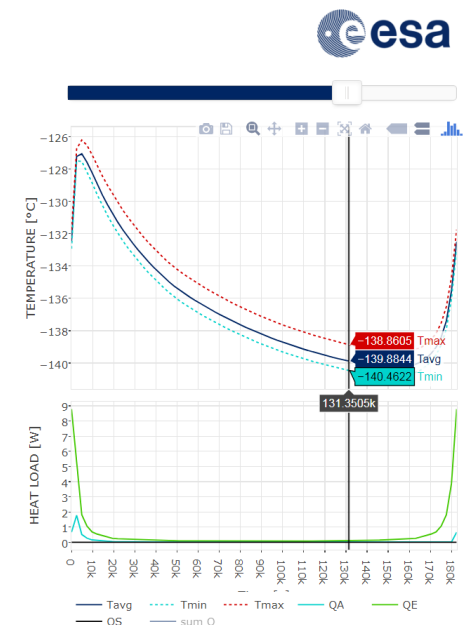


## Heat Flux Report

Results Report Info

Group: SXI Radiator

Group: SXI Radiator			
Time: 131350.54 [s]		Search...	
Heat from group	Conductive [W]	Radiative [W]	Total [W]
ENVIRONMENT		-4.0386e+0	-4.0386e+0
PLM Radiator mX		3.5435e-5	3.5435e-5
PM		2.6341e-1	2.6341e-1
SVM MLI		7.5918e-3	7.5918e-3
SVM Radiators		1.5412e-3	1.5412e-3
SXI FPA	3.2988e+0		3.2988e+0
SXI Radiator MLI	1.7698e-1	7.7041e-2	2.5402e-1
Solar Array		9.7598e-5	9.7598e-5
QA			7.3087e-3
QE			9.6728e-2
QS			1.8684e-2
Total [W]	3.4758e+0	-3.6888e+0	-9.0307e-2

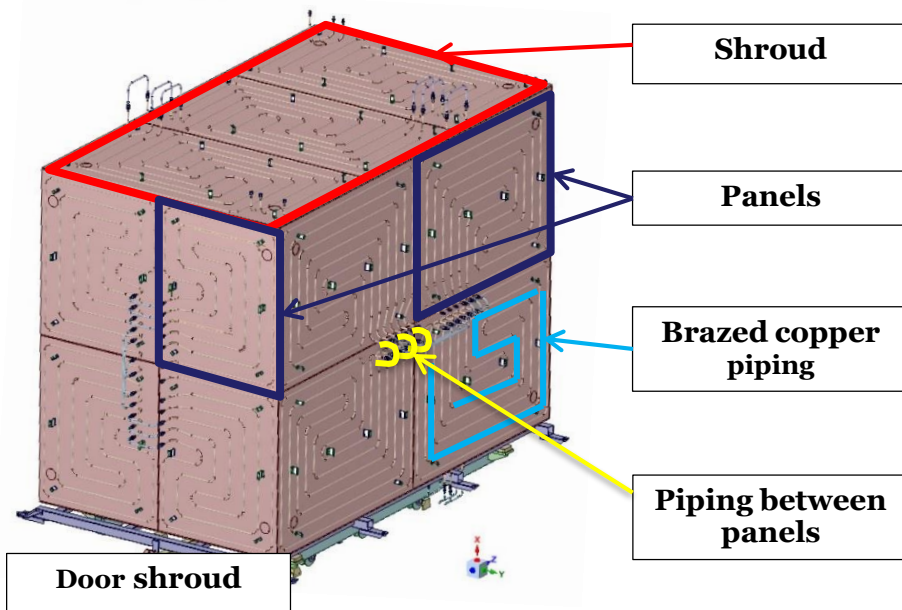




# Overview



- PHENIX Thermal Vacuum Facility
- Objectives
- Software selection
- ESATAN Model
  - Software description
  - Model description
  - Main heat path + convection
- EcosimPro Model
  - Software description
  - Model description
- Co-simulation
- User Interface
- Conclusions and model validation plan



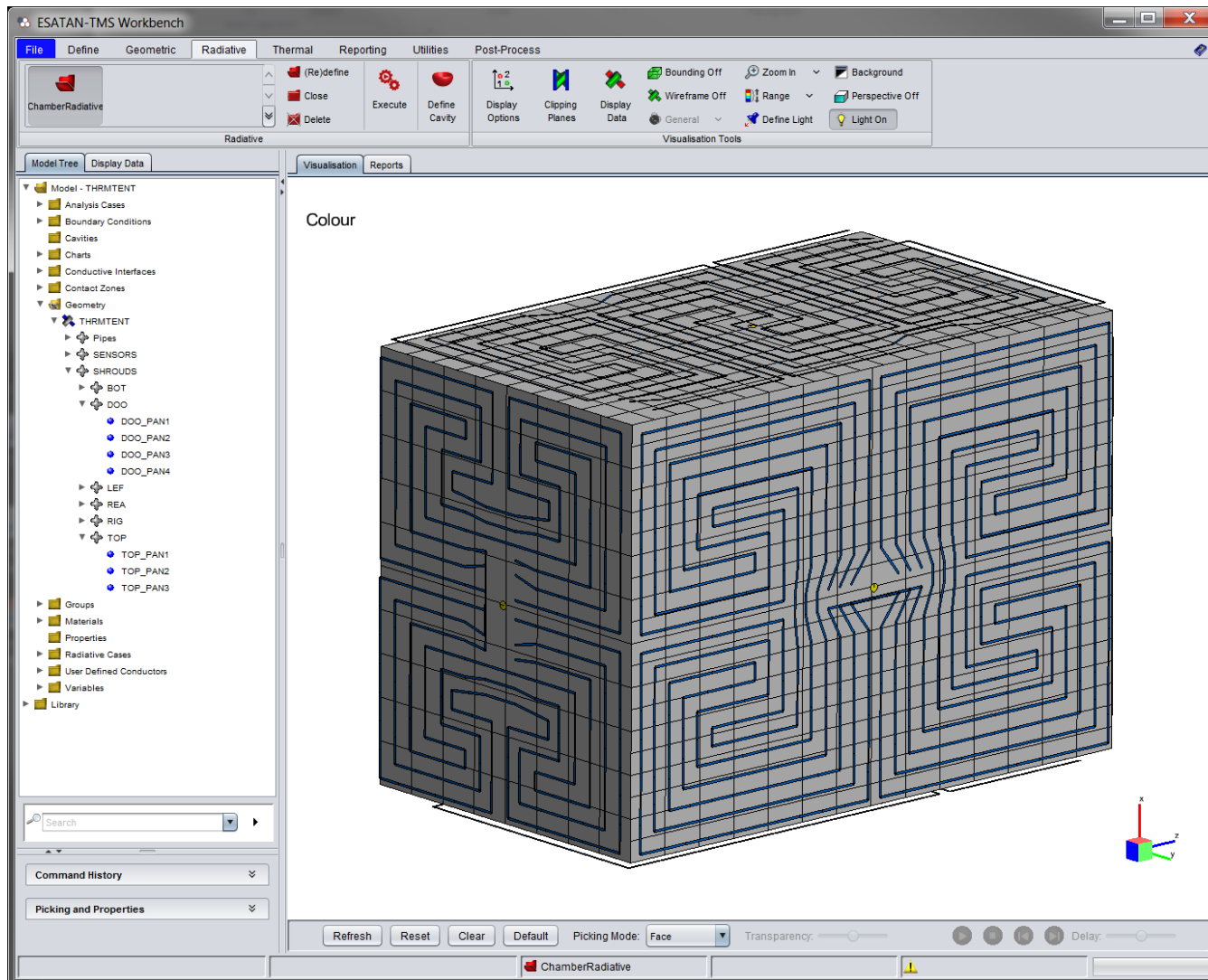
Name	Volume	Length	Diameter	Min Temp.	Max Temp.	Thermal Channels	Sun Diameter
VTC 1.5	10 m <sup>3</sup>	2.5 m	1.5 m	100 K	423 K	2	N/A
Phenix	160 m <sup>3</sup>	10 m	4.5 m	100 K	373 K	6	N/A
LSS	2300 m <sup>3</sup>	10 m	9.3 m	100 K	350 K	2	6 m



# Objectives



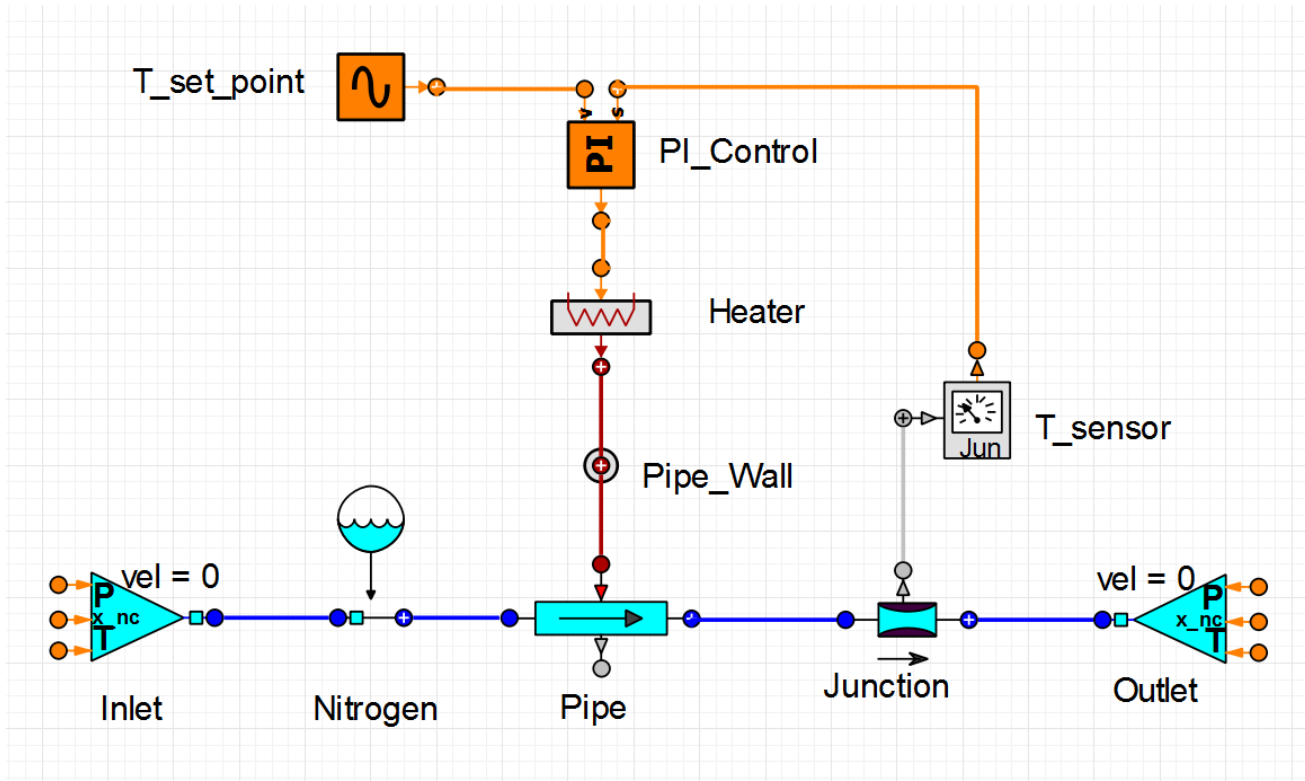
1. Design a feasibility test with a given chamber blockage and dissipation, determine the heat loads on each shroud:
  - $\text{GN}_2 = 600 \text{ W/shroud} - \text{max } 2\text{kW total}$
  - $\text{LN}_2 = 60 \text{ kW total}$
2. Estimate the shroud temperature homogeneity ( $\Delta T$ ), to determine if it is acceptable for a given test.
3. Develop a tool for use by a thermal non-specialist.





## Libraries:

- European Space Propulsion Simulation System (ESPSS)
- Co-simulation with ESATAN



- ESATAN-TMS

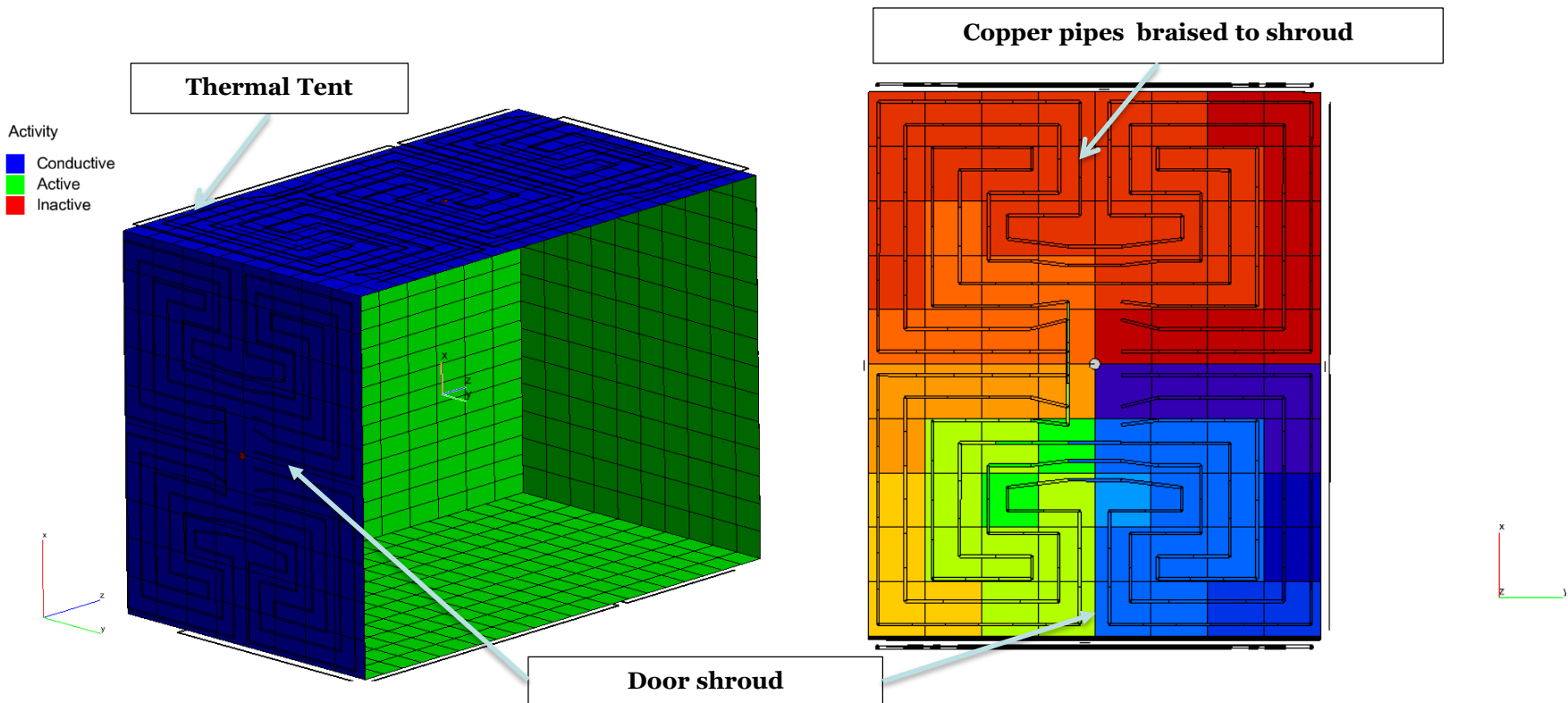
- Option to include a user's spacecraft thermal model
- Detailed temperature homogeneity map of each shroud

- EcosimPro

- Allow component modelling of pipes, valves, inlets, outlets and control PI logic – easy diagnostics and variable access
- Co-simulation library compatible with ESATAN-TMS
- Modelling of mixing of cold and hot GN<sub>2</sub> lines
- Option to integrate the tool into an Excel plug-in






- Radiative heat exchange inside the thermal tent
- Conductive heat exchanges between the pipe network and thermal tent walls
- Convective heat exchange from the fluid to pipe network






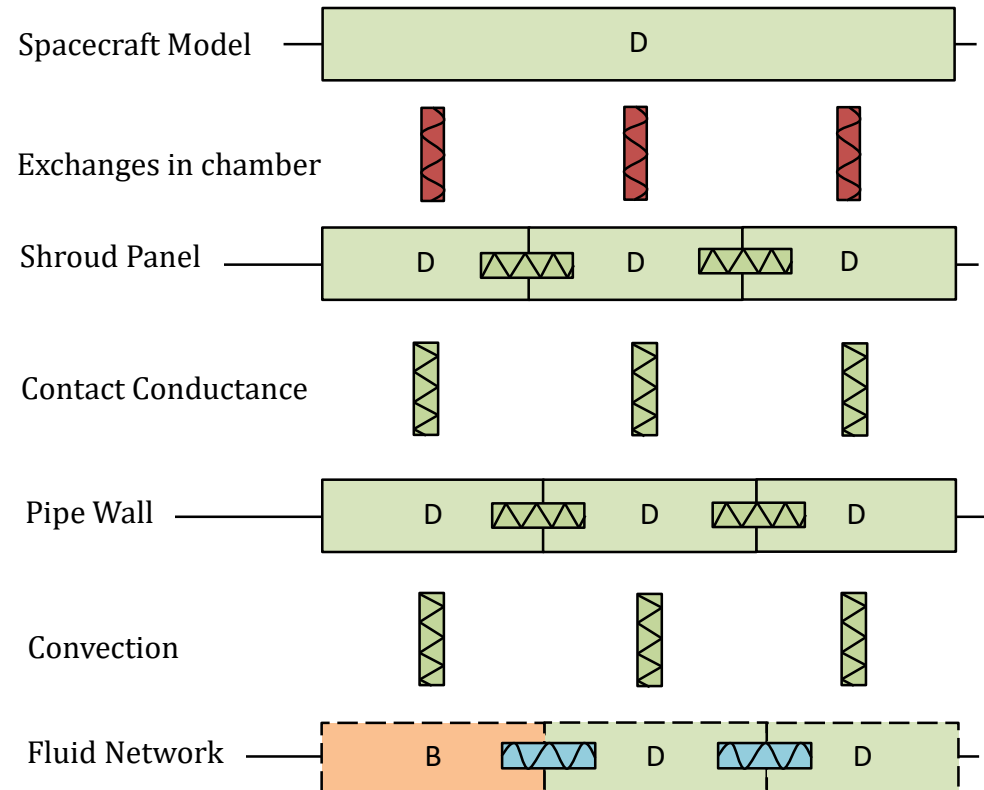
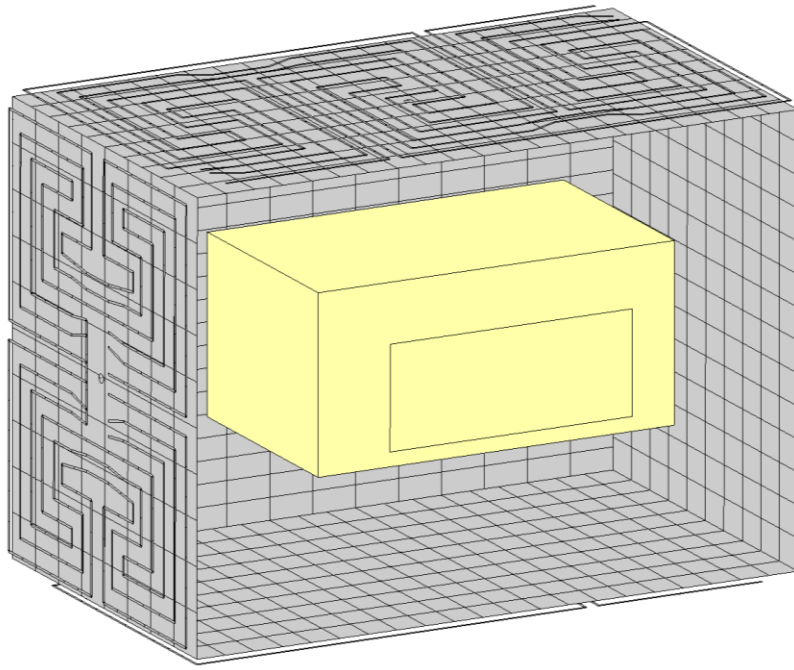
## Key

### Nodes

	Boundary
	Diffusion
	Diffusion - fluid

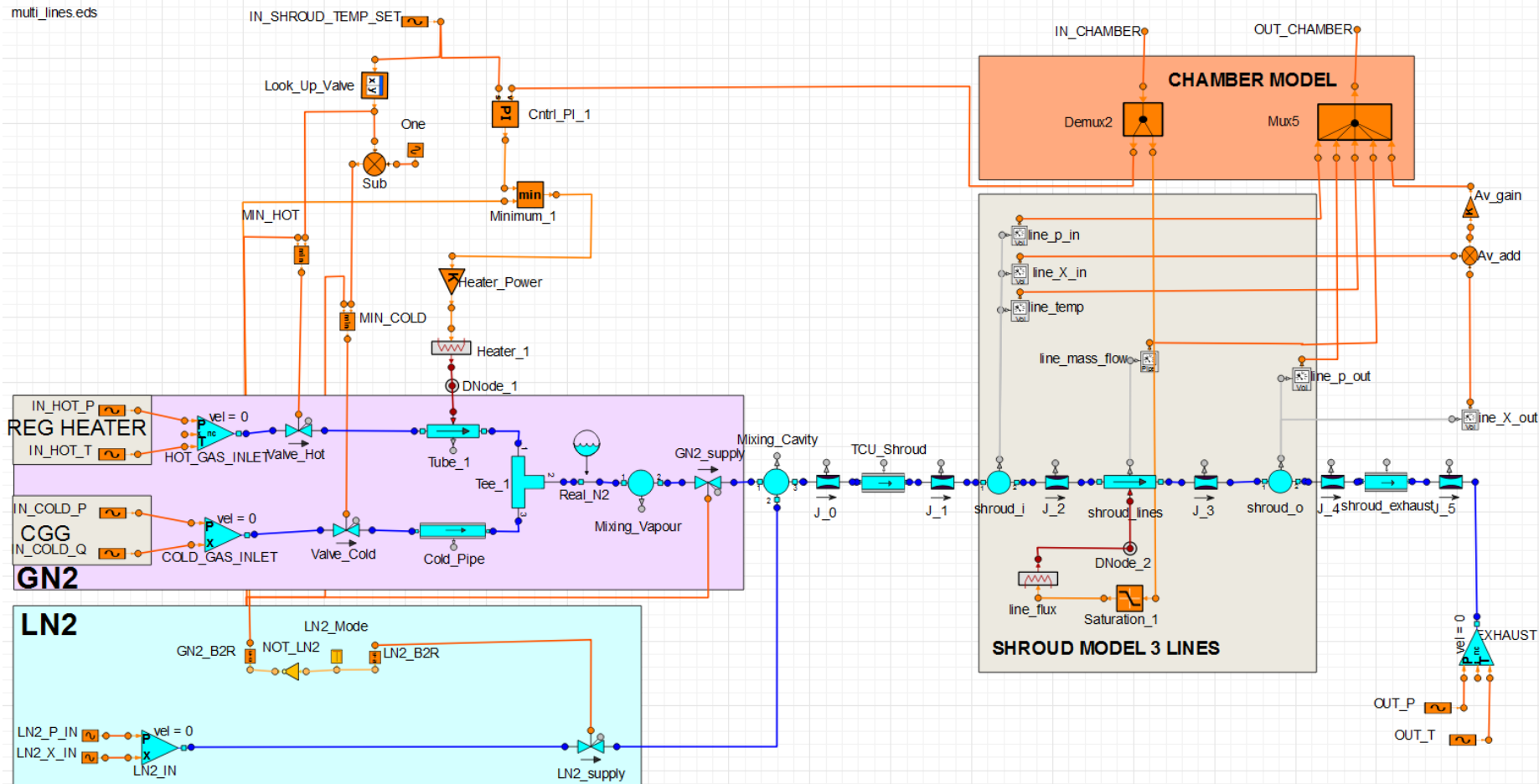
### Conductors

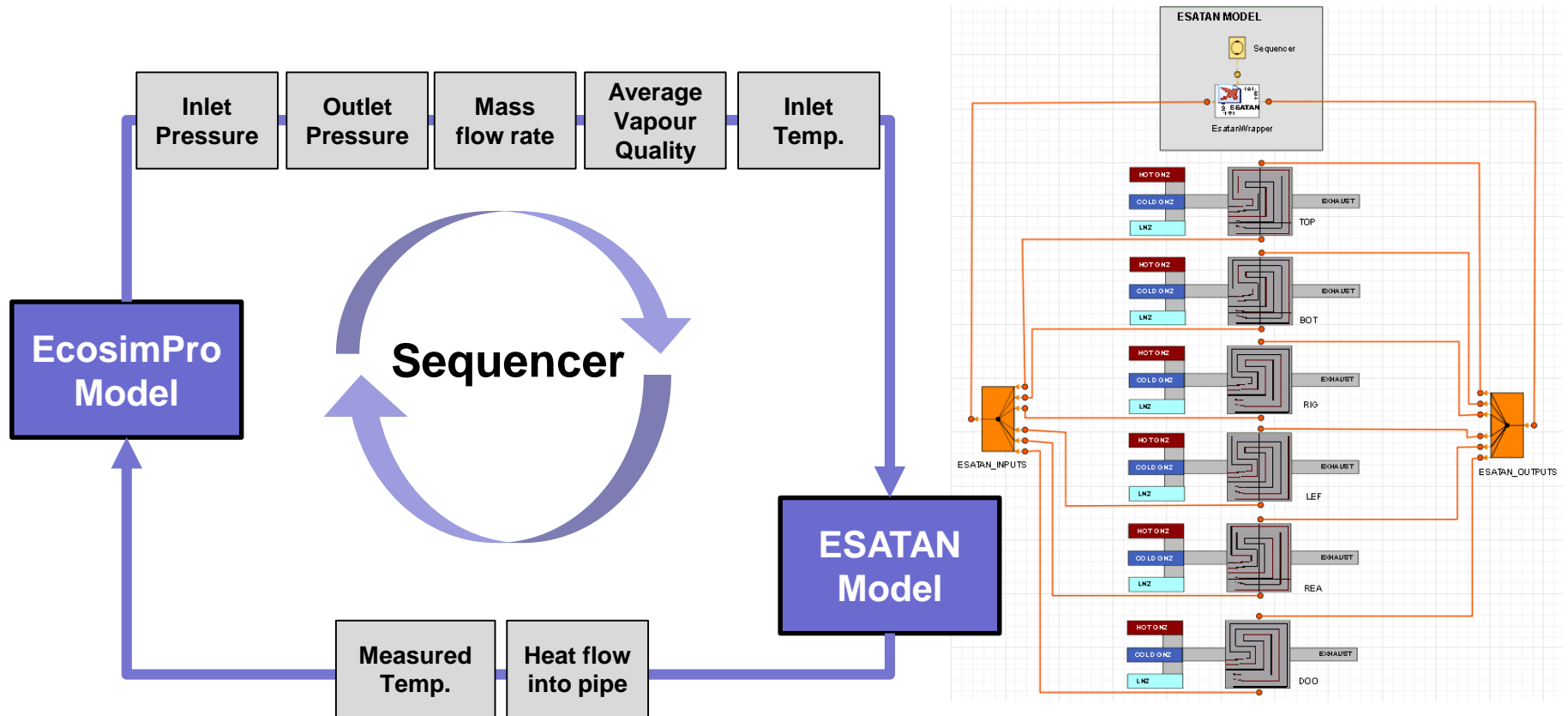
	GR: Radiative
	GF: Fluid
	GL: Linear



# Single Shroud Model

multi\_lines.ed5





Open

Close

Play

Pause

Stop

Add watch

Add report sheet

Remove watch

Clear all watches

$f_x$

New integration

New steady calculation

Save state

Restore state

Reset model

Reset selected watches

Settings

Add-in help

About

Help

Experiment file

Simulation

Experiment variables

Experiment customization

AC41

A

B

C

D

E

F

G

H

I

J

K

L

M

N

O

P

Q

R

S

T

U

V

W

X

Y

Z

A

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31

32

33

34

35

36

37

38

39

40

PHENIX CHAMBER SIMULATION TOOL

INLET HOT GN2

PRESSURE

300000

Pa

TEMP

293

K

INLET COLD GN2

PRESSURE

300000

Pa

QUALITY

1

INLET LN2

PRESSURE

200000

Pa

QUALITY

0

K

OUTLET

PRESSURE

101325

Pa

TEMP

293

K

PHENIX SHROUDS

Set Temp

LN, Mode

BACK

310

FALSE

BOTTOM

300

FALSE

DOOR

290

FALSE

NORTH

280

FALSE

SOUTH

190

FALSE

TOP

240

FALSE

SHROUD HEAT FLUX LIMIT

2000

W

BLOCKAGE GEOMETRY

X Length

2.7

m

X Translation

0

m

Y Length

1.2

m

Y Translation

0

m

Z Length

1.1

m

Z Translation

0

m

BLOCKAGE HEAT LOADS

BACK

0

W

BOTTOM

0

W

DOOR

0

W

NORTH

0

W

SOUTH

0

W

TOP

0

W

SHROUD TEMPERATURES

SHROUD HEAT FLUXES

Simulation output

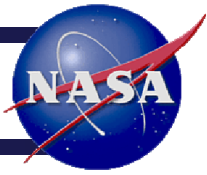
```

[TIME: 2580] Integration period 43
[TIME: 2640] Integration period 44
[TIME: 2665.95224] Detected new events (PHNX_LINE_SOUTH.Cntrl_PI_1.v[1] < PHNX_LINE_SOUTH.Cntrl_PI_1.u_min)
[TIME: 2700] Integration period 45
[TIME: 2760] Integration period 46
[TIME: 2820] Integration period 47
[TIME: 2880] Integration period 48
[TIME: 2940] Integration period 49
[TIME: 3000] Integration period 50
End of transient 1 (Status: OK, Runtime: 120.1s, Jacob.: 270, Residues: 16291)

```



# Model Validation



- Access to a range of thermocouple data on each shroud to estimate shroud temperature homogeneity
- Gaseous  $N_2$  valve positions can be adjusted in the model
- Exercise to correlate steady state set-points of pipe temperatures.

- [1] ESPSS European Space Propulsion System Simulation, EcosimPro Libraries User Manual, (VOLUME 1), 30-08-2015, Empresarios Agrupados, Madrid, Spain
- [2] ESATAN-TMS Thermal Modelling Suite, ITP Engines, Leicester, UK, Online: <https://www.esatan-tms.com/>, Accessed: 28-07-2017
- [3] ESTEC Test Centre Virtual Tour, ESA, Online: <http://esamultimedia.esa.int/multimedia/ESTEC/virtualtour/>, Accessed: 28-07-2017
- [4] CoolProp – User-friendly interface around the full capabilities of NIST REFPROP, Online: <http://www.coolprop.org/>, Accessed: 28-07-2017